

SUPPLEMENTAL TYPE CERTIFICATE

10060944 REV. 1

This Certificate/Approval is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation and in accordance with Commission Regulation (EU) No. 748/2012 to

KAMAN CORPORATION AIRCRAFT WHEEL AND BRAKE, LLC

**1160 CENTER ROAD
AVON OH 44011
UNITED STATES OF AMERICA**

and certifies that the change in the type design for the product listed below with the limitations and conditions specified meets the applicable Type Certification Basis and, if applicable, environmental protection requirements when operated within the conditions and limitations specified below:

Type Certificate Number: EASA.IM.A.277

Type Certificate Holder: TEXTRON AVIATION INC.

Type: A100, 200, 300, 1900

Model: 300

B300

B300C

Original STC Number: FAA STC SA380CH

Description of Design Change:

Replace existing nose wheel with Cleveland Nose Wheel P/N 40-204 by installing Parker Hannifin Conversion Kit 199-126

Revision 1 is an administrative re-issuance following a name change from Parker Hannifin Corporation Aircraft Wheel and Brake Division to Kaman Corporation, Aircraft Wheel and Brake, LLC.

EASA Certification Basis:

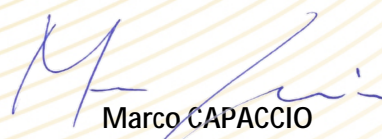
The Certification Basis (CB) for the original product remains applicable to this certificate/ approval.

The requirements for environmental protection and the associated certified noise and/ or emissions levels of the original product are unchanged and remain applicable to this certificate/ approval.

See Continuation Sheet(s)

For the European Union Aviation Safety Agency

Cologne, Germany, 07 June 2023



Marco CAPACCIO
Section Manager

Small Aircraft, Balloons & Airships



Associated Technical Documentation:

- Parker Hannifin Conversion Kit 199-126, Revision M, dated 31 July 2011 or later revisions of the above listed document(s) approved/accepted on behalf of EASA in accordance with the Technical Implementation Procedures of EU/ USA Bilateral Agreement.

Limitations/Conditions:

Prior to installation of this design change it must be determined that the interrelationship between this design change and any other previously installed design change and/ or repair will introduce no adverse effect upon the airworthiness of the product.

- End -